

XF01L
V 6.97

-2.0
-1.5
 C_p
-1.0
-0.5
0.0
0.5
1.0

BaseVent04

$Re = 2.000 \times 10^6$

$\alpha = 0.5587^\circ$

$C_L = 0.1500$

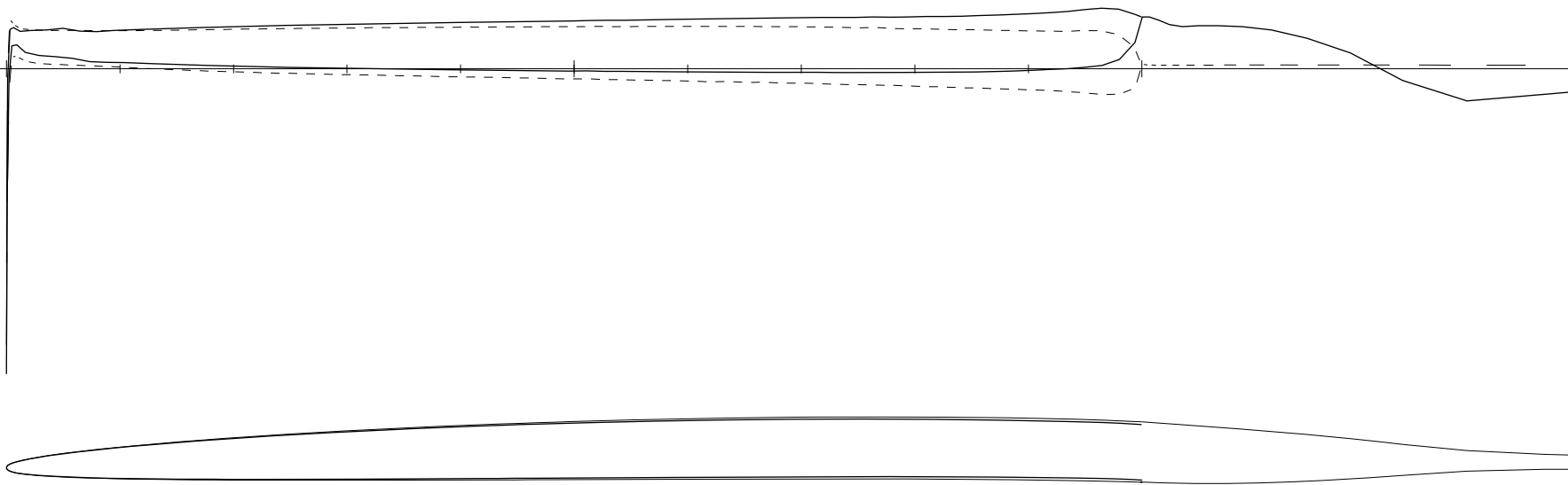
$C_M = -0.0447$

$C_D = 0.01558$

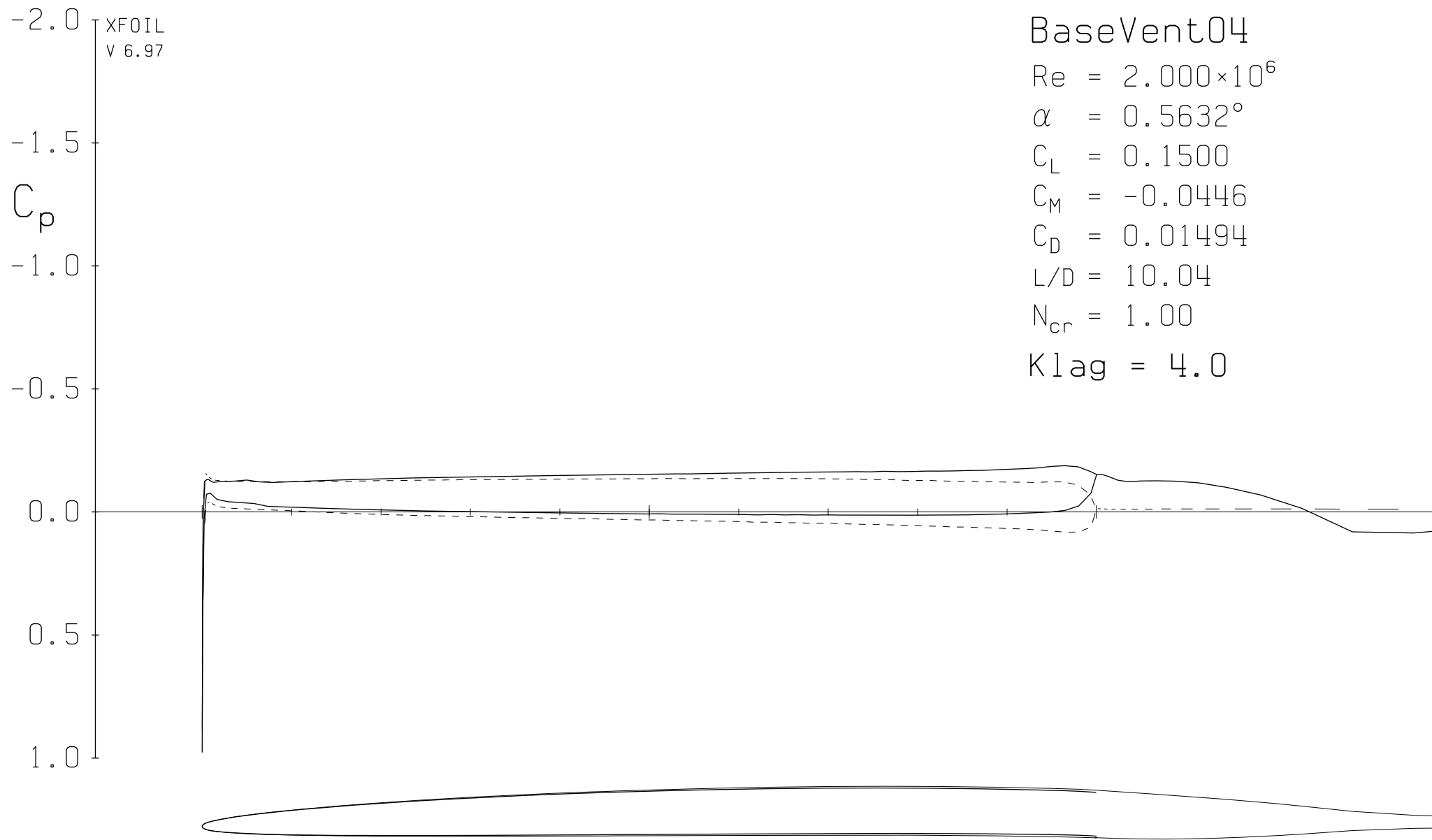
$L/D = 9.63$

$N_{cr} = 1.00$

Klag=5.6 (default)



XF0IL
V 6.97



BaseVent04

$Re = 2.000 \times 10^6$

$\alpha = 0.5632^\circ$

$C_L = 0.1500$

$C_M = -0.0446$

$C_D = 0.01494$

$L/D = 10.04$

$N_{cr} = 1.00$

$K_{lag} = 4.0$

XF01L
V 6.97

-2.0
-1.5
 C_p
-1.0
-0.5
0.0
0.5
1.0

BaseVent04

$Re = 2.000 \times 10^6$

$\alpha = 0.6303^\circ$

$C_L = 0.1500$

$C_M = -0.0426$

$C_D = 0.00811$

$L/D = 18.49$

$N_{cr} = 1.00$

$K_{lag} = 0.9$

