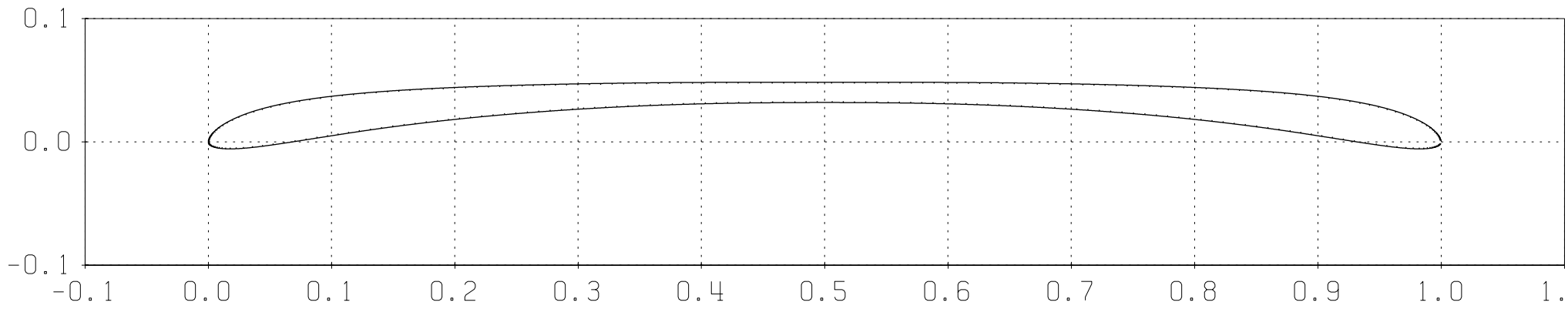
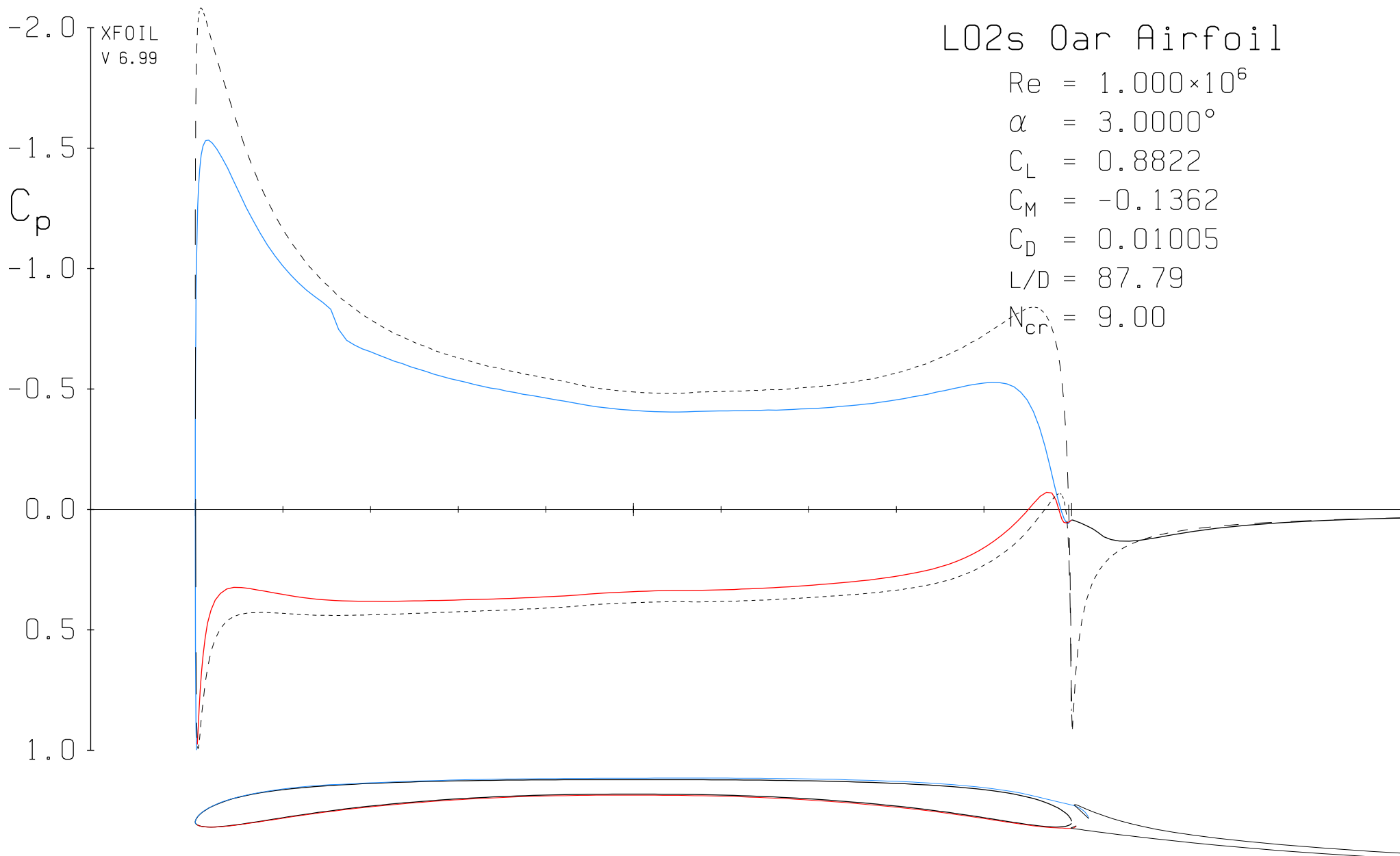


L02s 0ar Air  
area = 0.02313  
thick. = 0.03250  
camber = 0.04031  
 $r_{LE}$  = 0.00487  
 $\Delta\theta_{TE}$  = 142.49°



XFoil  
V 6.99



L02s Oar Airfoil

Re =  $1.000 \times 10^6$

$\alpha = 3.0000^\circ$

$C_L = 0.8822$

$C_M = -0.1362$

$C_D = 0.01005$

L/D = 87.79

$N_{cr} = 9.00$