

-2.0 XF01L
V 6.96

-1.5
 C_p

-1.0

-0.5

0.0

0.5

1.0

Segel 9%

$Re = 1.208 \times 10^6$

$\alpha = 8.5260^\circ$

$C_L = 1.5711$

$C_M = -0.1200$

$C_D = 0.02644$

$L/D = 59.43$

$N_{cr} = 0.50$

