

-2.0 xFOIL
v 6.97

-1.5

-1.0

C_p

-0.5

0.0

0.5

1.0

MRC16

$Re = 0.150 \times 10^6$

$\alpha = 8.5000^\circ$

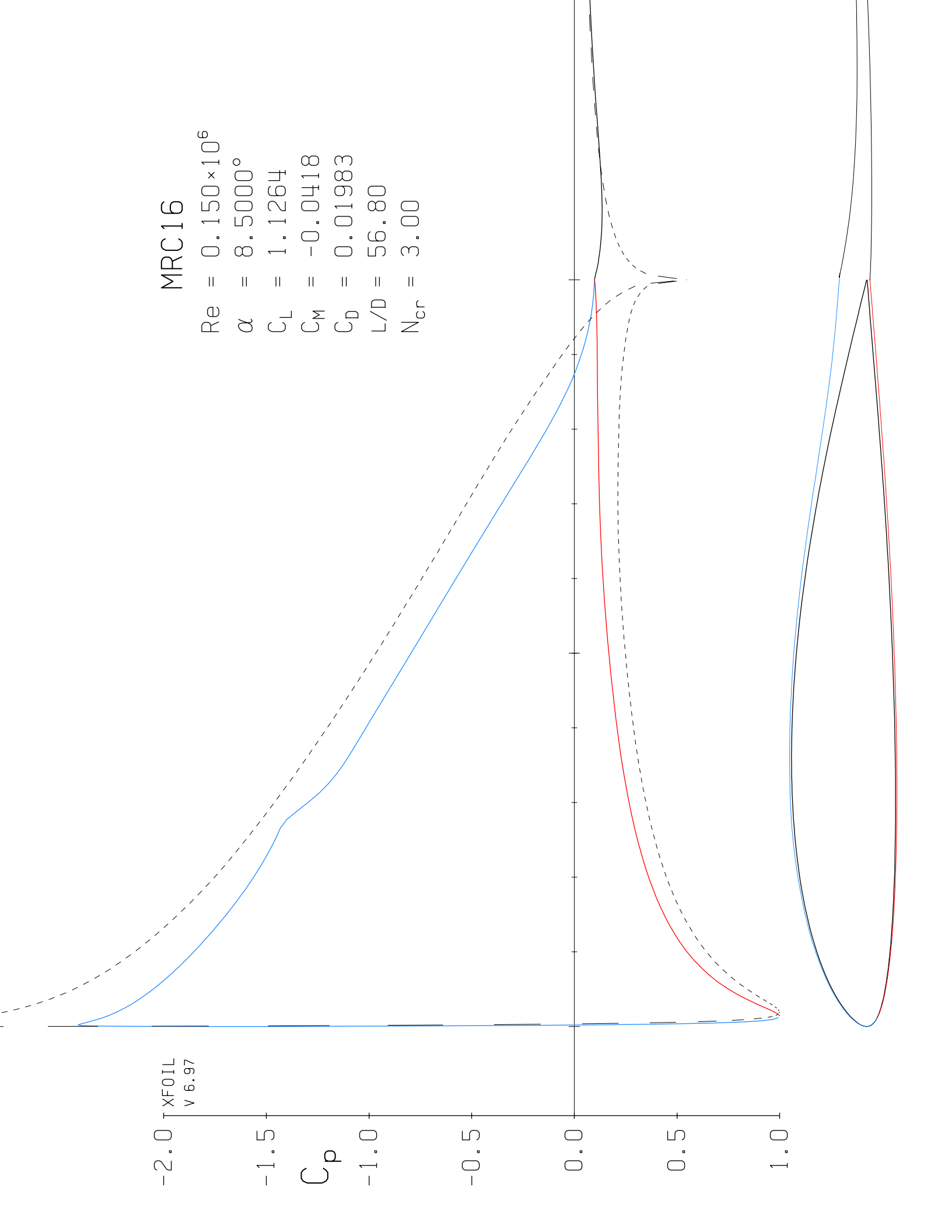
$C_L = 1.1264$

$C_M = -0.0418$

$C_D = 0.01983$

$L/D = 56.80$

$N_{cr} = 3.00$



-2.0 XF01L
V 6.97

MRC16

$Re = 0.300 \times 10^6$

$\alpha = -0.5000^\circ$

$C_L = 0.2658$

$C_M = -0.0620$

$C_D = 0.00912$

$L/D = 29.15$

$N_{cr} = 3.00$

C_p

