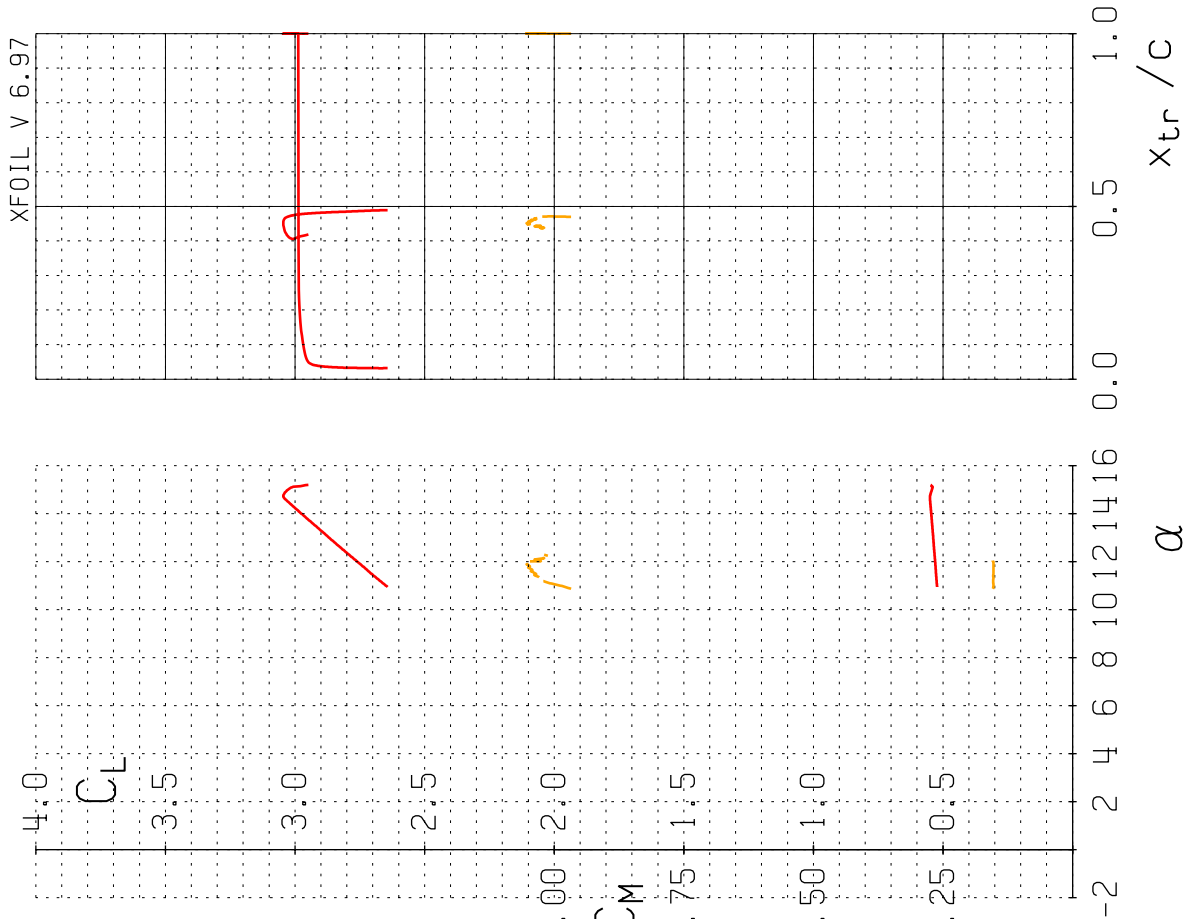
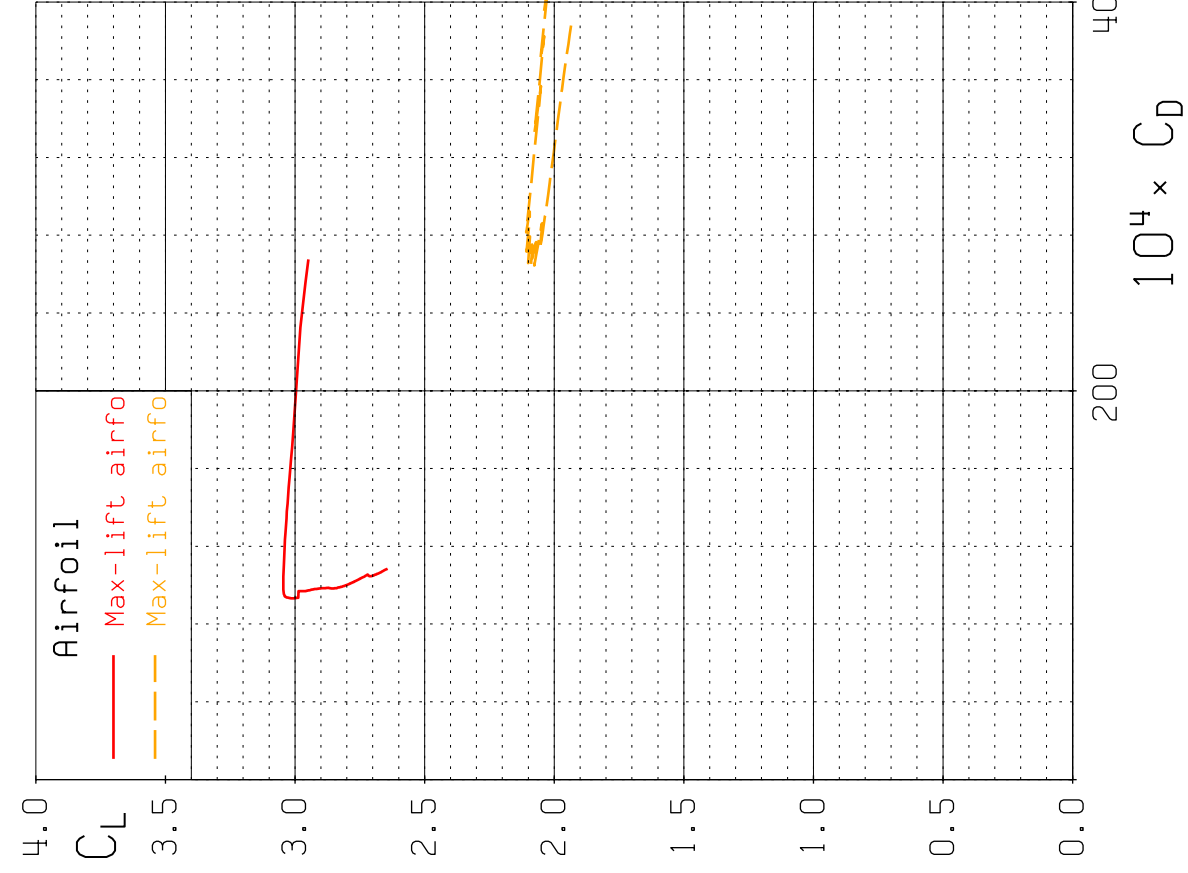


Max-lift airfoil 1 Re = 5000000 Ma = 0.000 Ncrit = 9.000
 Max-lift airfoil 5 Re = 300000 Ma = 0.000 Ncrit = 9.000



x_{tr} / c