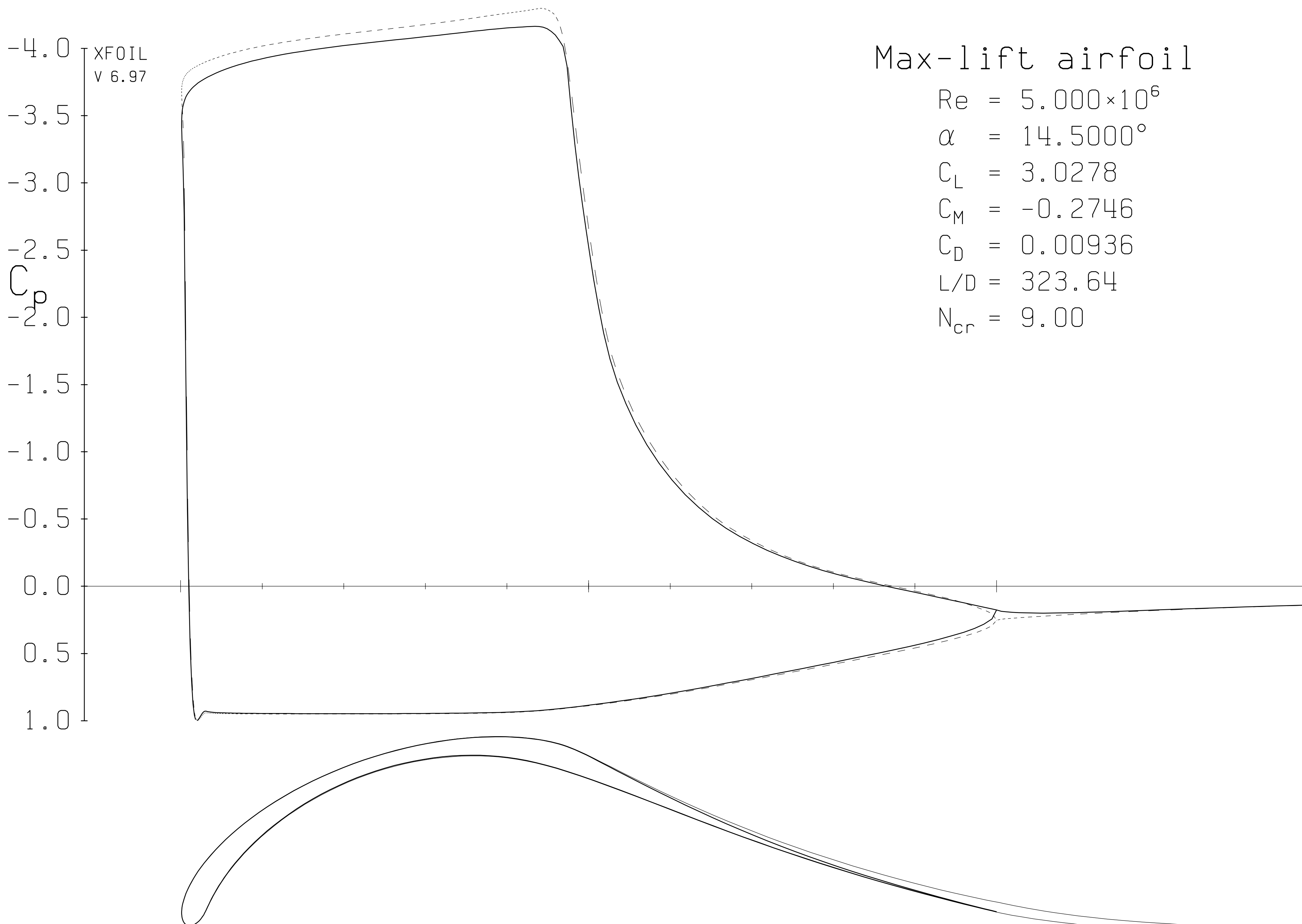
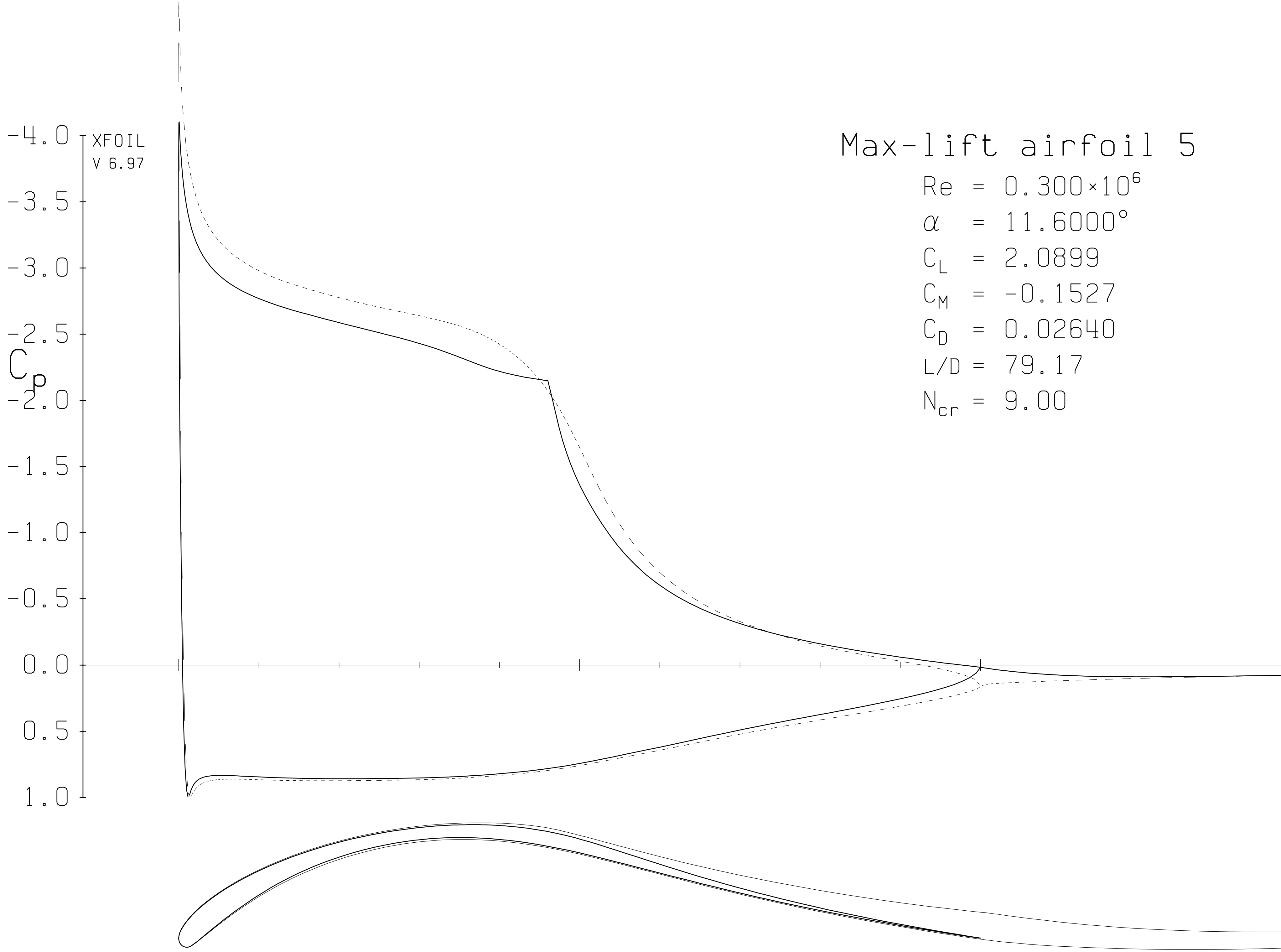


Max-lift airfoil
Re = 5.000×10^6
α = 14.5000°
C_L = 3.0278
C_M = -0.2746
C_D = 0.00936
L/D = 323.64
N_cr = 9.00
XFOIL
V 6.97
C_p
-4.0
-3.5
-3.0
-2.5
-2.0
-1.5
-1.0
-0.5
0.0
0.5
1.0



Max-lift airfoil 5
Re = 0.300×10^6
α = 11.6000°
C_L = 2.0899
C_M = -0.1527
C_D = 0.02640
L/D = 79.17
N_cr = 9.00
XFOIL
V 6.97
C_p
-4.0
-3.5
-3.0
-2.5
-2.0
-1.5
-1.0
-0.5
0.0
0.5
1.0